

B. M. S. College of Engineering, Bengaluru - 560019

Autonomous Institute Affiliated to VTU

August 2024 Supplementary Examinations

Programme: B.E.

Branch: Aerospace Engineering

Course Code: 20AE5DCBFM

Course: Basic Flight Mechanics

Semester: V

Duration: 3 hrs.

Max Marks: 100

Instructions: 1. Draw figures/expressions wherever necessary.
2. Assume suitable data wherever necessary.

UNIT - I

1. a) Define endurance. Derive an expression for endurance for a jet driven aircraft. 10
- b) Propeller driven aircraft flies at lower altitude, justify the statement. Also mention the necessary conditions for maximum range and endurance of jet propelled aircraft without changing the fuel capacity. 05
- c) Consider the Northrop F-5 fighter airplane, which has a wing area in 16 m^2 and wing span of 7.7 m . The wing is generating 80068 N of lift. For a flight velocity of 112 m/s at standard sea level, Calculate the lift co-efficient, induced drag co-efficient and induced drag. (Assume $e = 0.8$) 05

OR

2. a) Derive an expression for turn radius and turn rate. 12
- b) Enumerate drag polar, mention the different types of drag polar. 08

UNIT - II

3. a) Enumerate the balance field length for take-off. 12
- b) Write short notes on V-n diagram. 8

UNIT - III

4. a) Derive an expression for tail contribution for longitudinal stability of an aircraft. 12
- b) Elucidate statically stable, unstable and neutrally stable system with an example. 08

Important Note: Completing your answers, compulsorily draw diagonal cross lines on the remaining blank pages. Revealing of identification, appeal to evaluator will be treated as malpractice.

OR

5. a) State the criteria for longitudinal static stability. Explain the same with an illustration. **12**
- b) Prove that the neutral point is the aerodynamic center of the complete airplane. **8**

UNIT - IV

6. a) Describe the illustration of aerodynamic moment of elevator – hinge moment. **10**
- b) Prove that the free elevator factor $F = 1 - \frac{1}{a_t} \frac{\partial C_{L_t}}{\partial \delta_e} \left(\frac{\partial C_{h_e} / \partial \alpha_t}{\partial C_{h_e} / \partial \delta_e} \right)$ **10**

UNIT - V

7. a) Explain main rotor & tail rotor **14**
- b) List the major components of helicopter, and list the differences between fixed wing aircraft and helicopter **06**
